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 $2 \pi L C = (3.4.4)$ Substituting (3.4.4)
into (3.4.2) we have, $2 \pi L C = (3.4.4)$

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We can find the optimum solution by $\frac{\partial \pi}{\partial a} = 0$, by some operations leads to $\frac{\partial \pi}{\partial a} = 0$,
therefore we have $a = \sqrt{\frac{2C}{b}}$,
Substitute it back to (3.4.4), we have $\pi = \frac{2}{3} \sqrt{\frac{2C}{b}}$ (3.4.5) That means the optimum cross-section for elliptical shapes is a circle.

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1.1 The beam of a rectangular thin-walled section (i.e., t is very small) is designed to carry both bending moment M and torque T . If the total wall contour length (see Fig. 1.16) is fixed, find the optimum b/a ratio to achieve the most efficient section if) ($2 b a L + = T M =$

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and allowable allowable $\tau \sigma 2 = .$

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structures C.T. Sun 2222) 2 (2 aCa L b

$-\text{---} = \pi$, where 2) 2 (2 $\pi L C = (3.4.4)$

Substituting (3.4.4) into (3.4.2) we have,

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22 aCaAellp $= \pi$ We can find the optimum solution by $0 = \partial \partial a A$, by some operations leads to $0 \ 2 \ 22 \ 22 = -$
 $- = \partial \partial aC \ aC \ a A \ \pi$, therefore

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() 06155. 2 2 2 3 4 2 1 2 2 = + + - = r,

The central coordinate of the circle is ()

0, 5. 3 2 2 2, 2 3 4, = |) \ | \ (- + = c

c y x Therefore we have the maximum

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and minimum stresses, respectively,
 $\sigma_{max} = \frac{\sigma_x + \sigma_y}{2} + r$
 $\sigma_{min} = \frac{\sigma_x + \sigma_y}{2} - r$
These are the same as we obtained above.

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Consider a state of stress ...

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(3) Two approaches are employed to find the solution. (i) Assume that the bending stress reaches the allowable allowable first and find the

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corresponding bending maximum bending moment. C T Sun Mechanics of Aircraft Structures Solution... C T Sun Mechanics of Aircraft Structures Solution.

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